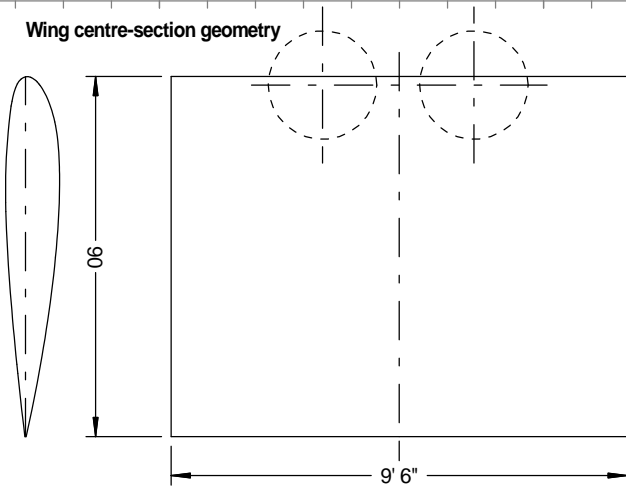


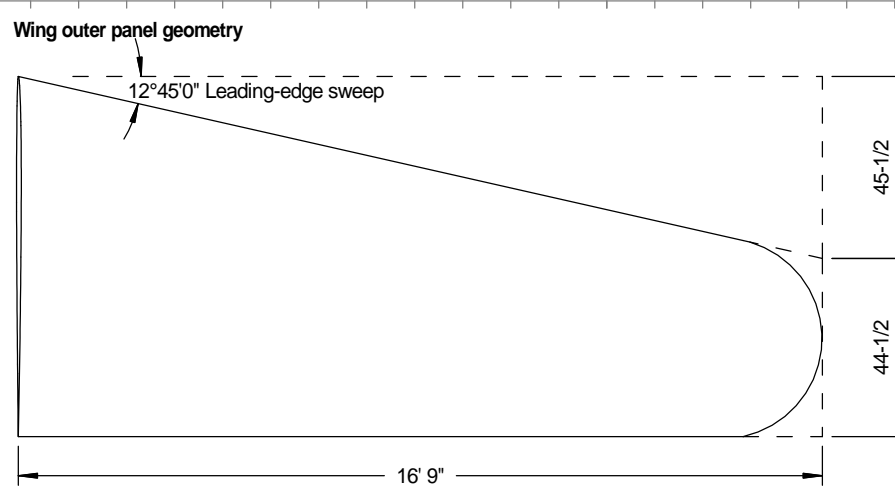
1 foot scale on drawing border

Wing centre-section geometry



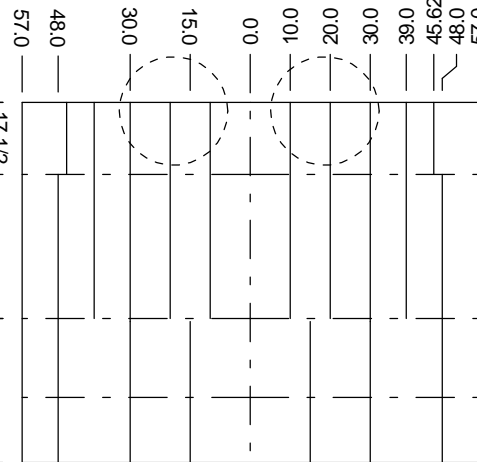
Centre-section airfoil:
 NACA 2215
 Maximum thickness: 15%
 Maximum thickness location: 30% of chord
 Maximum camber: 2% of chord
 Maximum camber location: 20% of chord
 Wing attached to fuselage at 2° incidence

Wing outer panel geometry

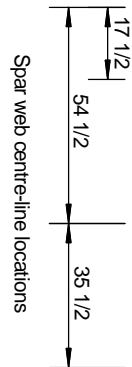


Airfoil at theoretical tip:
 NACA 4412
 Maximum thickness: 12% of chord
 Maximum thickness location: 30% of chord
 Maximum camber: 4% of chord
 Maximum camber location: 40% of chord
 Washout: 2°0'

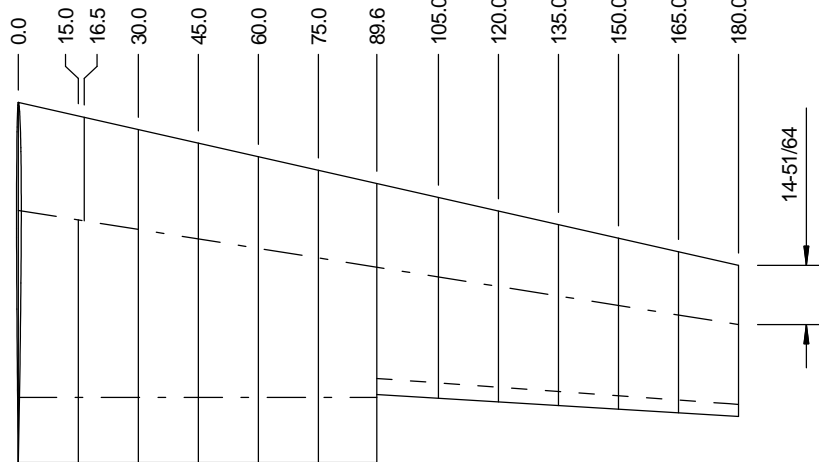
Wing centre-section stations



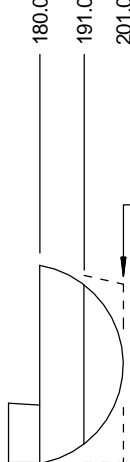
Spar web centre-line locations



Wing outer panel stations



Wing tip stations



Theoretical tip (station 201)

- References:
 1. Wirraway Overhaul and Repair Manual; Commonwealth Aircraft Corporation, 1940; RAAF Publication 76
 2. Measurements taken from A20-10 at Australian National Aviation Museum, Moorabbin, Victoria

Chord-plane of centre section

Chord-plane of outer panel

