



Airfoil at root:
 NACA 2215
 Thickness: 15%
 Maximum camber: 2% of chord
 Maximum camber location: 20% of chord
 Wing attached to fuselage at 2° incidence

Airfoil at theoretical tip:
 NACA 4412
 Thickness: 12% of chord
 Maximum camber: 4% of chord
 Maximum camber location: 40% of chord
 Washout: 2°

References:
 1. Wirraway Overhaul and Repair Manual; Commonwealth Aircraft Corporation, 1940; RAAF Publication 76
 2. Sketch by Peter Malone based on research by Richard Hourigan

Note: Features and dimensions shown in red are scaled from drawings and need to be confirmed with actual measurements